

loading, we know that one half of the external loads at joints U_1 and L_1 is supported at reaction R_1 and $1/2$ at reaction R_2 , or shear in



center panel = $(100 - 50) 1/2 = 75$. The vertical shear in panel $U_1U_2L_1L_2$ equals 75 plus the external loads at U_1 and L_1 or a total of 225 and similarly for the end panel shear = $225 + 50 + 100 = 375$. With these values known, the other two sides of the index triangles are directly proportional to the sides of the length triangles for each panel, and the results are as shown in Fig. A2.30.

The general procedure from this point is to find the loads in the diagonals, then in the verticals, and finally in the horizontal chord members.

The loads in the diagonals are equal to the values on the hypotenuse of the index triangles. The sense, whether tension or compression, is determined by inspection by cutting mental sections thru the truss and noting the direction of the external shear load which must be balanced by the vertical component of the diagonals.

The loads in the verticals are determined by the method of joints and the sequence of joints is so selected that the stress in the vertical member is the only unknown in the equation $\Sigma V = 0$ for the joint in question.

Thus for joint U_1 , $\Sigma V = -50 - U_1L_1 = 0$ or $U_1L_1 = -50$.

For joint U_2 , $\Sigma V = -50 - U_2L_2 - U_1L_1 = 0$, but $U_1L_1 = 75$, the vertical component of U_1L_1 from index triangle. $\therefore U_2L_2 = -50 - 75 = -125$. For joint L_1 , $\Sigma V = -100 - L_1U_1 = 0$, hence $L_1U_1 = 100$.

Since the horizontal chord members receive their loads at the joints due to horizontal components of the diagonal members of the truss, we can start at L_1 and add up these horizontal components to obtain the chord stresses. Thus, $L_1L_2 = 312$ (from index triangle). $L_2L_3 = 312$ from $\Sigma H = 0$ for joint L_2 . At joint U_1 , the

horizontal components of diagonal members = $312 + 187.8 = 499.8$. Therefore, load in $U_1U_2 = -499.8$. Similarly at joint L_2 , $L_2L_3 = 312 + 187.8 = 499.8$. At joint U_2 , the horizontal components of U_1U_2 and $U_2U_3 = 499.8 + 52.3 = 552$ which must be balanced by -552 in member U_2U_3 .

The reaction R_2 equals the value on the vertical side of our index triangle in the end panel, or 375. This should be checked using the truss as a whole and taking moments about R_1 .

If a truss is loaded unsymmetrically, the reactions should be determined first, after which the index triangles can be drawn, starting with the end panels, since the panel shear is then readily calculated.

A2.12 Aircraft Wing Structure. Truss Type with Fabric or Plastic Cover

The metal covered cantilever wing with its better overall aerodynamic efficiency and sufficient torsional rigidity has practically replaced the externally braced wing except for low speed commercial or private pilot aircraft as illustrated by the aircraft in Figs. A2.31 and 32. The wing covering is usually fabric and therefore a drag truss inside the wing is necessary to resist loads in the drag truss direction. Figs. A2.33 and 34 show the general structural layout of such wings. The two spars or beams are metal or wood. Instead of using double wires in each drag truss bay, a single diagonal strut capable of taking either tension or compressive loads could be used. The external brace struts are stream line tubes.



Fig. A2.31 Piper Tri-Pacer

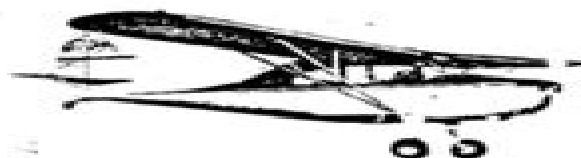



Fig. A2.32 Champion Traveler

Engineering Analysis Of Flight Vehicles

Harvey H. Hubbard



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